Project 2 – MANE 6710 Numerical Design Optimization

*Unique ID: 5284*

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# Introduction

The main objective of this project is to design the cross-sectional profile of a spar of an aircraft wing. The idea is to arrive at a design which leads to the least weight possible for the structure given certain design parameters. The optimal design that comes as a result should not fail under the loading conditions and in fact it must be manufacturable.

# Choice of Design Variable

With the cross section fixed to be a circular annulus, the most obvious choice of design variables are the Radii of the annulus along the length of the spar. The parameterization of design variable chosen in this project is as follows:

The length of the spar can be discretized into elements () and with a pair of design variables at each node as shown in Figure 1.

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Figure . Cross- sectional Profile of the spar

# Objective Function and Constraints

The objective of this project is to optimize the design of the spar’s cross-section that has the least weight possible.

This cross-sectional profile needs to adhere to certain design parameters such as:

1. Semi Span Length –
2. Shape of Cross section – Circular Annulus
3. Material – Carbon Fiber composite – {Density ;Young’s modulus ; Yield Compressive/Tensile stress }
4. Aircraft operational weight –
5. Loading – with force distribution linearly varying along the length (maximum load at the root and no load at the tip)

With these parameters, the objective function will be the Mass or the Weight of the spar which can be calculated as . Hence this design problem can now be formulated as:

|  |  |
| --- | --- |
|  | Equation 1 |

This minimization problem is subject to constraints that stem from the manufacturing side of the spar, and they are:

1. along the entire length of the spar
2. along the entire length of the spar
3. along the entire length of the spar
4. along the entire length of the spar

MATLAB’s in-built optimizer is used to perform the optimization of spar cross-section.

## Calculation of Volume

Volume is generally calculated by the integral where, is the cross-section along the length of the spar. With very fine discretization of the length of spar into , this integral can be approximated as:

|  |  |
| --- | --- |
|  | Equation 2 |

This calculates the average area of the cross-section on two adjacent node points and multiplies it with the distance between these two node points . This is a variational form of trapezoidal integration technique.

## Calculation of Constraints

### Setting up Linear Inequality constraints

Constraints are set up as Linear Inequality constraints of the format requires it in - where is the matrix that contains the linear mapping between the design variables and constraint values .

### Setting up Nonlinear constraints

This is the most important aspect of this project as the computation of normal stress at each cross-section requires the use of numerical methods like Finite Element Methods to calculate it. The Area moment of Inertia for each cross-section along the length is,

|  |  |
| --- | --- |
|  | Equation 3 |

Then, the force which varies linearly along the length of the wing is calculated at each Node point as

|  |  |
| --- | --- |
|  | Equation 3 |

The operational load considered in this problem is Operational Weight -

Using the Area Moment of Inertia and the force on each node, the vertical displacement and the angular displacement of the wing at each node is calculated using the FEM (). Then, it gets easier to solve for the Normal stress along the length of the spar. This is calculated using function. Most important aspect of any optimization problem is to perform optimization in scales that are the same for all the parameters that are involved in this process. Since, the calculated maximum at each node is at large scales compared to the value of the objective function or the design variables, this has been normalized and the final nonlinear inequality constraint is written as

|  |  |
| --- | --- |
|  | Equation 4 |

# Setting up Optimization Problem

With the objective and the constraints set up, it is now important to calculate accurate gradients of the objective and the constraints for successful optimization of the design. Complex step gradient method is used to compute the gradients of the objective and the inequality constraints. With a very small complex step ,the round off errors caused by the central difference approximation can be avoided.

Two functions, and are used to compute the gradients of the objective function and the nonlinear constraint function.

Number of elements - 100

Initial point chosen -

Options chosen for –

1. Choice of Algorithm – ‘Sequential Quadratic Programming (SQP)’
2. Gradients – Provided and computed using Complex Step Algorithm

Initial weight of the spar was

Initial Design choice is shown in Figure 2

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Figure . Initial Design Parameterization

After optimization is performed, the optimal profile of the cross-section is shown in Figure 3

Chart, line chart

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Figure . Optimal design of Spar Cross-section

Chart, line chart

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Figure . First Order Optimality condition

Figure 4 shows the first order optimality condition to decrease and this conveys the fact that the optimizer has converged to a local minimizer which also satisfies the constraints.

The final weight of this optimal design is **.** This is a reduction of weight from the initial design.

# Results

A convergence analysis was performed to find if the mass estimate computed in this project improves as the number of elements are increased. As more cross-sections are added along the length elements, approximate volume calculated using the trapezoidal integral method improves as decreases and a small increase in the mass computed can be noticed as shown in Figure 5.

Chart, line chart

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Figure 5. Mass estimates (v) number of elements

Optimized profile undergoes the following deformation, stress under the application of a linearly varying force along the length of the semi spar of the wing as shown in Figure 6.

Chart

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Figure 6. Optimized Spar Performance

This profile for the spar not only has decreased its mass but has also adhered to all the manufacturing constraints specified for the design of the spar. This is the final optimized profile achieved using this parameterization for the choice of design variables .

# Appendix

The MATLAB code written for this project has been attached in this section.

















